## Model Question Paper - 1 with effect from 2020-21(CBCS Scheme)

USN

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## Fifth Semester B.E. Degree Examination FLIGHT MECHANICS

TIME: 03 Hours
Max. Marks: 100
Note: 01. Answer any FIVE full questions, choosing at least ONE question from each MODULE.

| Module - 1 |  |  | Marks |
| :---: | :---: | :---: | :---: |
| Q. 1 | (a) | With a neat sketch explain forces \& moments of an aircraft in the plain of symmetry. | 10 |
|  | (b) | Derive an expression for Cmcg for an aircraft | 10 |
| OR |  |  |  |
| Q. 2 | (a) | With a explain steady state performance of an aircraft | 10 |
|  | (b) | Write a note on contribution of airframe components for longitudinal stability. | 10 |
| Module - 2 |  |  |  |
| Q. 3 | (a) | Explain the estimation of hinged moment parameter for longitudinal stability | 10 |
|  | (b) | Write a note on stick free gradient in un accelerated flight. | 10 |
| OR |  |  |  |
| Q. 4 | (a) | Discuss about the difference between stick fixed \& stick free condition for longitudinal stability. | 4 |
|  | (b) | Derive an expression for neutral point of static longitudinal stability. | 12 |
|  | (c) | Write a note on trim tabs | 4 |
| Module - 3 |  |  |  |
| Q. 5 | (a) | Derive an expression for contribution of airframe component of directional stability. | 12 |


|  | (b) | Obtain the minimum control speed in the event of an engine failure for the following airplane. $S=65 \mathrm{~m}^{2}, \mathrm{~Sv}=6.5 \mathrm{~m}^{2}, \mathrm{Lv}=10.5 \mathrm{M}, \mathrm{BHP}=880 \mathrm{KW}$, Propeller efficiency=75\%, $\mathrm{Yp}=4.5 \mathrm{M},\left(\mathrm{dC}_{\mathrm{Lv}} / \mathrm{d} \delta \mathrm{v}\right)=0.02 / \mathrm{deg},(\delta \mathrm{r}) \max =25 \%$. | 8 |
| :---: | :---: | :---: | :---: |
| OR |  |  |  |
| Q. 6 | (a) | Derive an expression for estimation of Dihedral effect for lateral stability | 10 |
|  | (b) | Explain the estimation of lateral control power of an aircraft. | 10 |
| Module - 4 |  |  |  |
| Q. 7 | (a) | Discussed about gravitational and thrust velocity equation of a flight. | 10 |
|  | (b) | Why deflecting the aileron produce a yawing moment, explain with a neat sketch. | 10 |
| OR |  |  |  |
| Q. 8 | (a) | Explain small perturbation theory \& its equation of motion. | 10 |
|  | (b) | Explain the following <br> i) Pitching velocity <br> ii) Time rate of change of an angle of attack. | $\begin{aligned} & \hline 5 \\ & 5 \end{aligned}$ |
| Module - 5 |  |  |  |
| Q. 9 | (a) | Explain Phegoid \& short period of motion for dynamic longitudinal stability | 10 |
|  | (b) | Examine the two potential cases where the Routh method breaks down as follows, <br> (a) $\lambda^{5}+\lambda^{4}+3 \lambda^{3}+3 \lambda^{2}+4 \lambda+6=0$ <br> (b) $\lambda^{6}+3 \lambda^{5}+6 \lambda^{4}+12 \lambda^{3}+11 \lambda^{2}+9 \lambda+6=0$. | 10 |
| OR |  |  |  |
| Q. 10 | (a) | Determine whether the characteristic equation given below have stable or unstable. <br> (a) $2 \lambda^{3}+6 \lambda^{2}+8 \lambda+8=0$ <br> (b) $2 \lambda^{3}+16 \lambda^{2}+4 \lambda+12=0$ <br> (c) $\mathrm{A} \lambda^{4}+\mathrm{B} \lambda^{3}+\mathrm{C} \lambda^{2}+\mathrm{D} \lambda+\mathrm{E}=0$. | 10 |
|  | (b) | With a neat sketch explain the Dutch roll and spiral mode of an aircraft. | 10 |



## Model Question Paper -2 with effect from 2020-21(CBCS Scheme)

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## Fifth Semester B.E. Degree Examination FLIGHT MECHANICS

TIME: 03 Hours
Max. Marks: 100
Note: 01. Answer any FIVE full questions, choosing at least ONE question from each MODULE.

| Module - 1 |  |  | Marks |
| :---: | :---: | :---: | :---: |
| Q. 1 | (a) | With a relevant graph, explain atmosphere and International standard atmosphere of an aircraft. | 10 |
|  | (b) | Derive an expression for (dcm/dcl) for various parts of an aircraft. | 10 |
| OR |  |  |  |
| Q. 2 |  | Explain power effect on static longitudinal stability of an aircraft. | 20 |
| Module - 2 |  |  |  |
| Q. 3 | (a) | Discussed about hinged parameter for longitudinal stability for stick-free condition. | 10 |
|  | (b) | Explain about restriction of aft C.G | 5 |
|  | (c) | Write a note on trim tabs | 5 |
| OR |  |  |  |
| Q. 4 | (a) | Briefly explain about floating characteristics and aerodynamic balance of static longitudinal stability. | 10 |
|  | (b) | Derive an expression for neutral point of static longitudinal stability. | 10 |
| Module - 3 |  |  |  |
| Q. 5 | (a) | Derive an expression for stick-free directional stability of a flight. | 10 |


|  | (b) | Discussed about rudder effect and dorsal fin. | 10 |
| :---: | :---: | :---: | :---: |
| OR |  |  |  |
| Q. 6 | (a) | Explain the following <br> i)swept wing <br> ii)flaps <br> iii)one engine inoperative condition | $\begin{aligned} & 4 \\ & 4 \\ & 4 \end{aligned}$ |
|  | (b) | Obtain the minimum control speed in the event of an engine failure for the following airplane. $S=65 \mathrm{~m}^{2}, S v=6.5 \mathrm{~m}^{2}, \mathrm{Lv}=10.5 \mathrm{M}, \mathrm{BHP}=880 \mathrm{KW}$, Propeller efficiency=75\%, $\mathrm{Yp}=4.5 \mathrm{M},\left(\mathrm{dC}_{\mathrm{Lv}} / \mathrm{d} \delta \mathrm{v}\right)=0.02 / \mathrm{deg},(\delta \mathrm{r}) \mathrm{max}=25 \%$. | 8 |
| Module - 4 |  |  |  |
| Q. 7 | (a) | Derive an expression of rigid body equations of motion | 20 |
| OR |  |  |  |
| Q. 8 | (a) | Explain small perturbation theory \& its equation of motion. | 10 |
|  | (b) | Explain the following i) Rolling rate <br> ii) Yawing rate | $\begin{aligned} & \hline 5 \\ & 5 \end{aligned}$ |
| Module - 5 |  |  |  |
| Q. 9 | (a) | Explain the derivative for roll-yaw coupling. | 10 |
|  | (b) | Examine the two potential cases where the Routh method breaks down as follows, <br> (a) $\lambda^{5}+\lambda^{4}+3 \lambda^{3}+3 \lambda^{2}+8 \lambda+6=0$ <br> (b) $\lambda^{6}+5 \lambda^{5}+6 \lambda^{4}+12 \lambda^{3}+11 \lambda^{2}+9 \lambda+6=0$. | 10 |
| OR |  |  |  |
| Q. 10 | (a) | Discussed about auto rotation and spin of an aircraft. | 10 |
|  | (b) | Determine whether the characteristic equation given below have stable or unstable. <br> (a) $\lambda^{3}+6 \lambda^{2}+12 \lambda+8=0$ <br> (b) $2 \lambda^{3}+4 \lambda^{2}+4 \lambda+12=0$ <br> (c) $\mathrm{A} \lambda^{4}+\mathrm{B} \lambda^{3}+\mathrm{C} \lambda^{2}+\mathrm{D} \lambda+\mathrm{E}=0$. | 10 |




